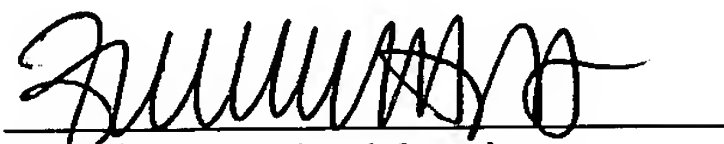


REMARKS

Please enter the foregoing preliminary amendment prior to examination of the present application. Claims 1-20 are now pending in this application. Applicants submit that this Amendment adds no new matter.

Respectfully Submitted,



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AMENDMENT TO THE SPECIFICATION:

Please replace paragraphs [0013] and [0014] with the following amended paragraphs:

[0013] Figure 1 is a schematic illustration of a gas turbine engine 10 including a high pressure compressor 14, and a combustor 16. Engine 10 also includes a high pressure turbine 18 and a low pressure turbine 20. Compressor 14 and turbine 18 are coupled by a first shaft 24, and turbine 20 drives a second output shaft 26. Shaft 26 provides a rotary motive force to drive a driven machine, such as, but, not limited to a gearbox, a transmission, a generator, a fan, or a pump. Engine 10 also includes a recuperator 28 that has a first fluid path 29 [30] coupled serially between compressor 14 and combustor 16, and a second fluid path 31 [32] that is serially coupled between turbine 20 and ambient 35 [34]. In one embodiment, the gas turbine engine is an LV100 available from General Electric Company, Cincinnati, Ohio. In an alternative embodiment, engine 10 includes a low pressure compressor 12 coupled by a first shaft 24 to turbine 20, and compressor 14 and turbine 18 are coupled by a second shaft 26.

[0014] In operation, air flows through high pressure compressor 14. The highly compressed air is delivered to recuperator 28 where hot exhaust gases from turbine 20 transfer heat to the compressed air. The heated compressed air is delivered to combustor 16. Airflow from combustor 16 drives turbines 18 and 20 and passes through recuperator 28 before exiting gas turbine engine 10. In an alternative embodiment, during operation, air flows through low pressure compressor 12 and compressed air is supplied from low pressure compressor 12 to high pressure compressor 14. The highly compressed air is delivered to combustor 16. Airflow from combustor 16 drives turbines 18 and 20 before exiting gas turbine engine 10.

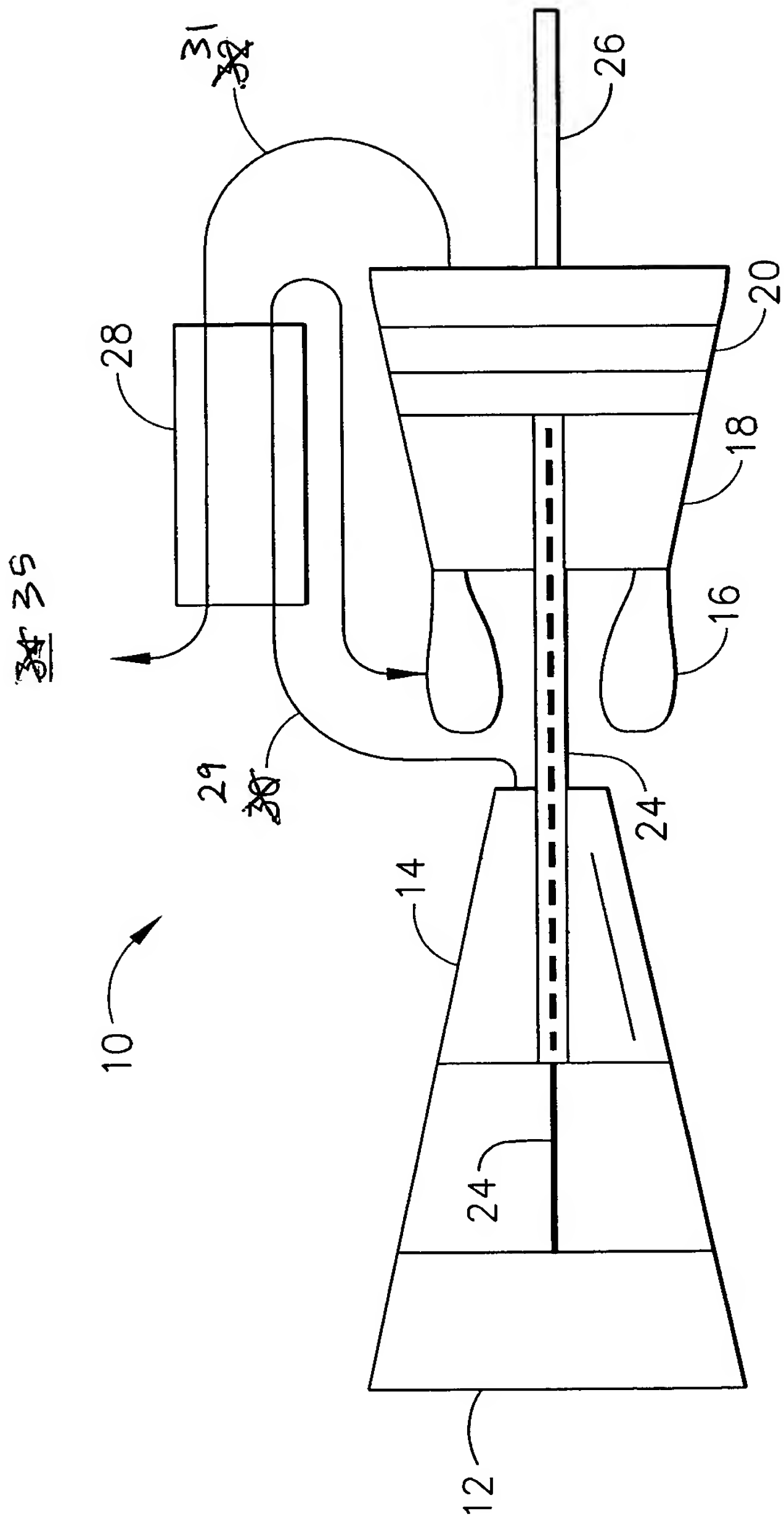


FIG. 1